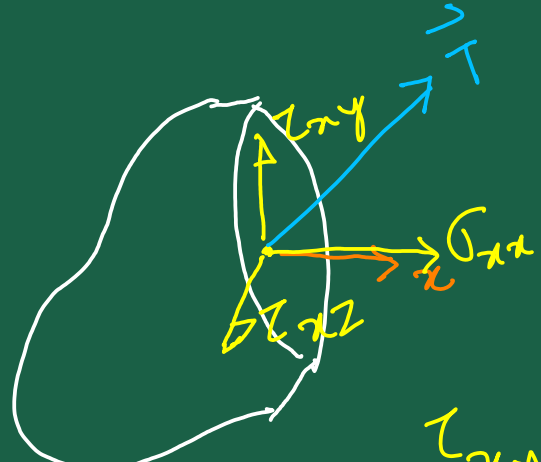
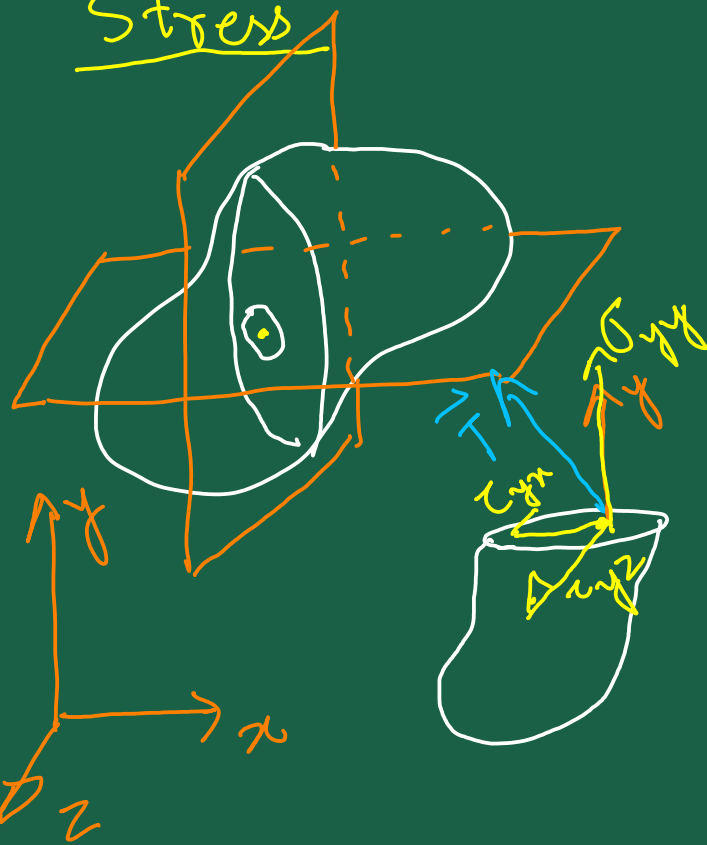


Stress

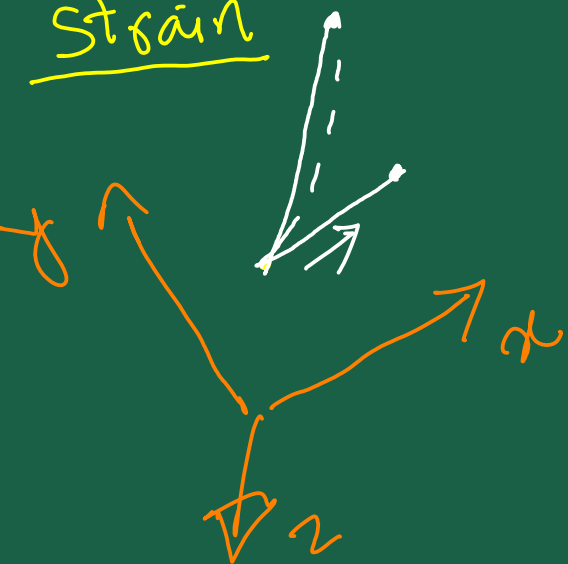


$$\lim_{A \rightarrow 0} \frac{\vec{F}}{A} = \vec{T}$$

$$\tau_{xy} = \tau_{yx}$$

$$\theta_1 \approx \tan \theta_1 = \text{ratio of lengths}$$

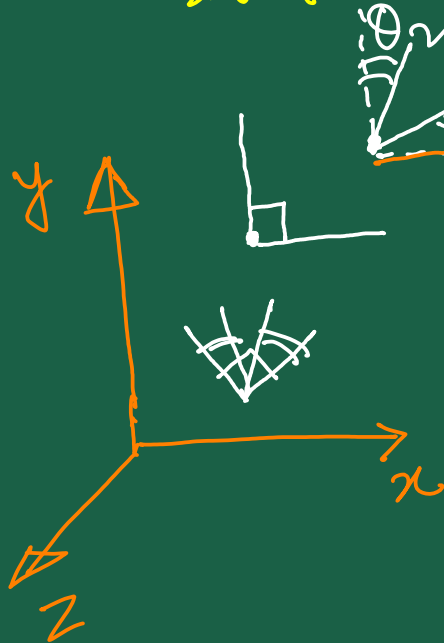
Strain



$$\epsilon_{xx} \uparrow \text{Normal strain}$$

$$\frac{DL}{L}$$

Shear Strain



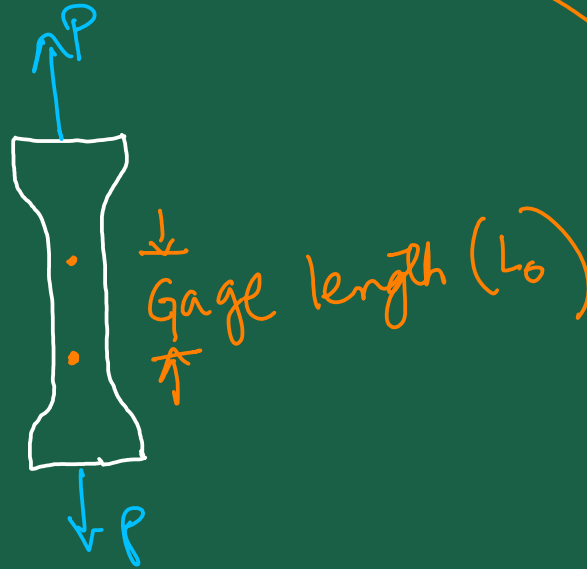
$$\theta_1 + \theta_2 = \gamma$$

shear strain

Nominal or engineering stress/strain

$$\sigma = \frac{P}{A_0}$$

Original area



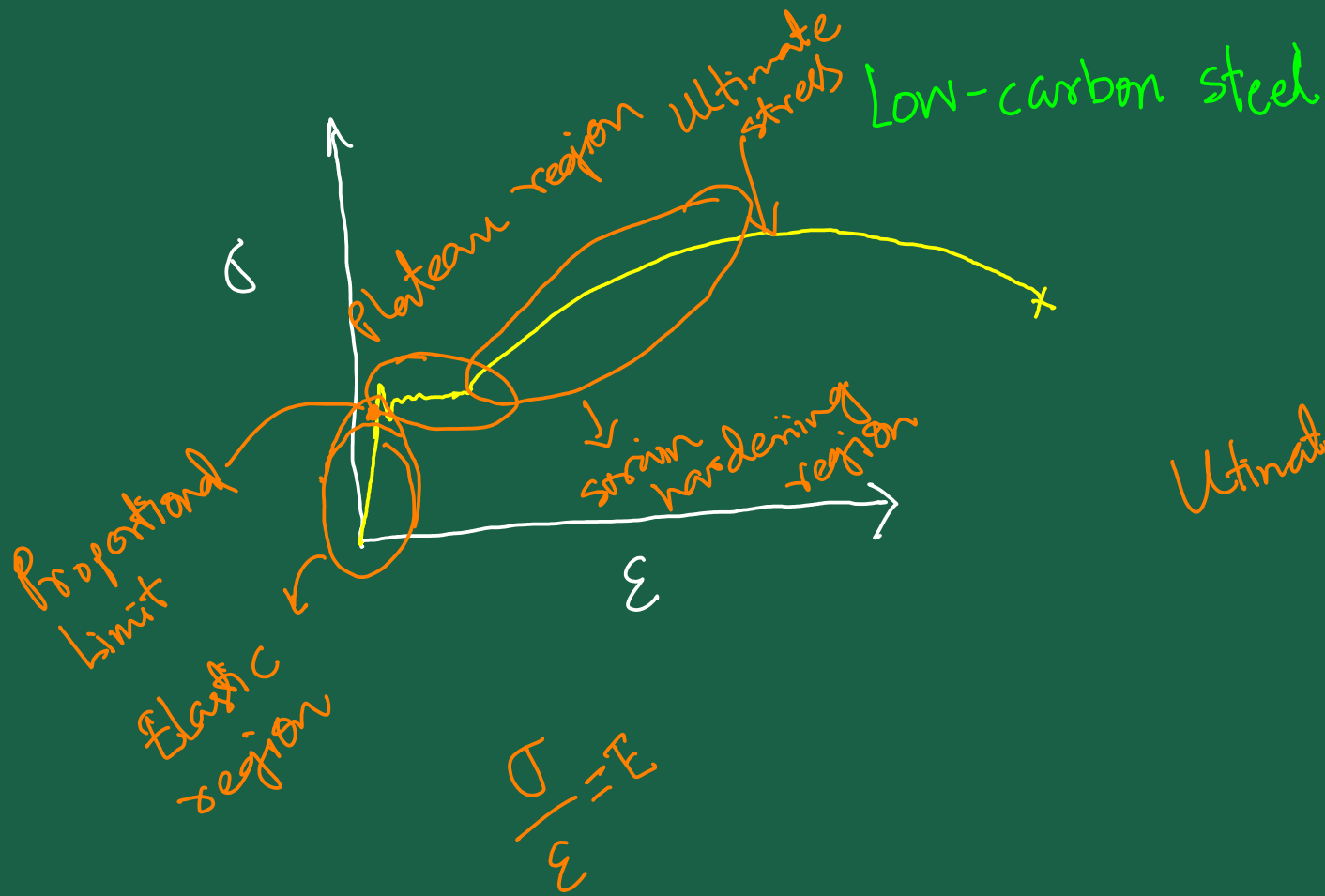
$$\frac{L - L_0}{L_0}$$

True stress/strain

$$\bar{\sigma} = \frac{P}{A}$$

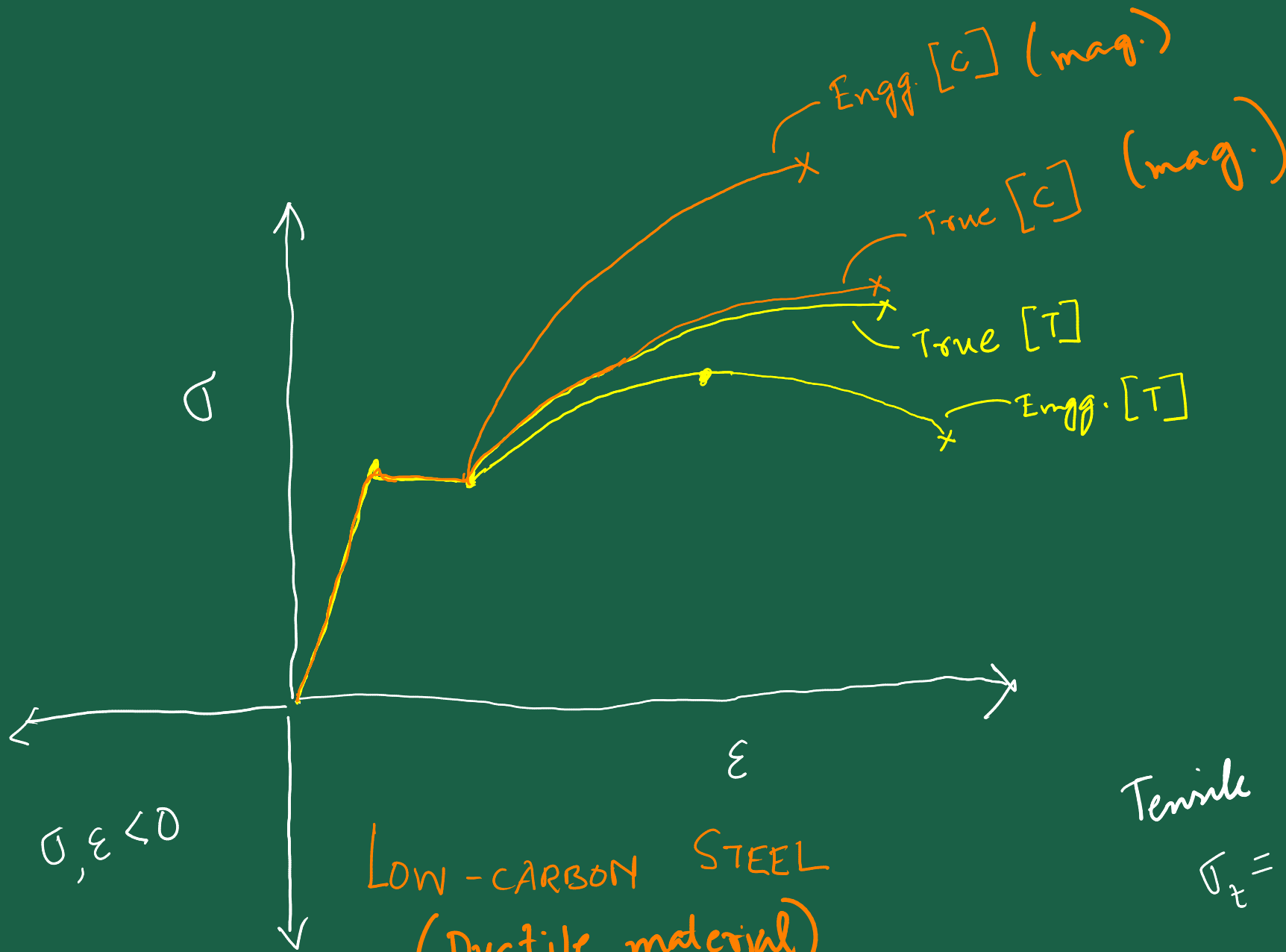
Current area

$$\epsilon = \int_{L_0}^{L_f} \frac{dL}{L} = \ln \frac{L_f}{L_0}$$



Ultimate stress \rightarrow Strength

Engineering stress - engineering strain diagram

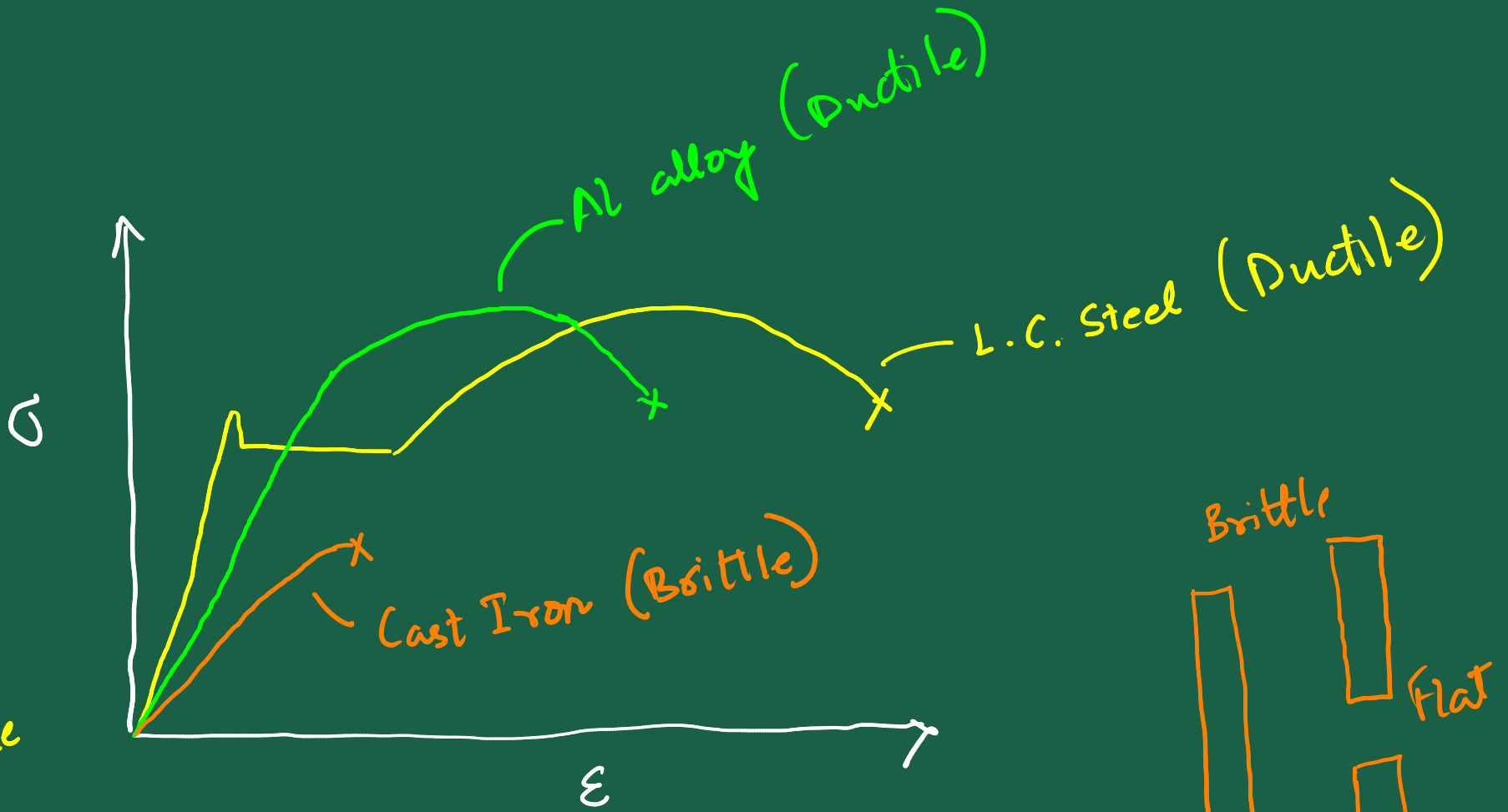


Tensile

$$\sigma_t = \frac{F}{A}$$

$$\sigma_e = \frac{F}{A_0}$$

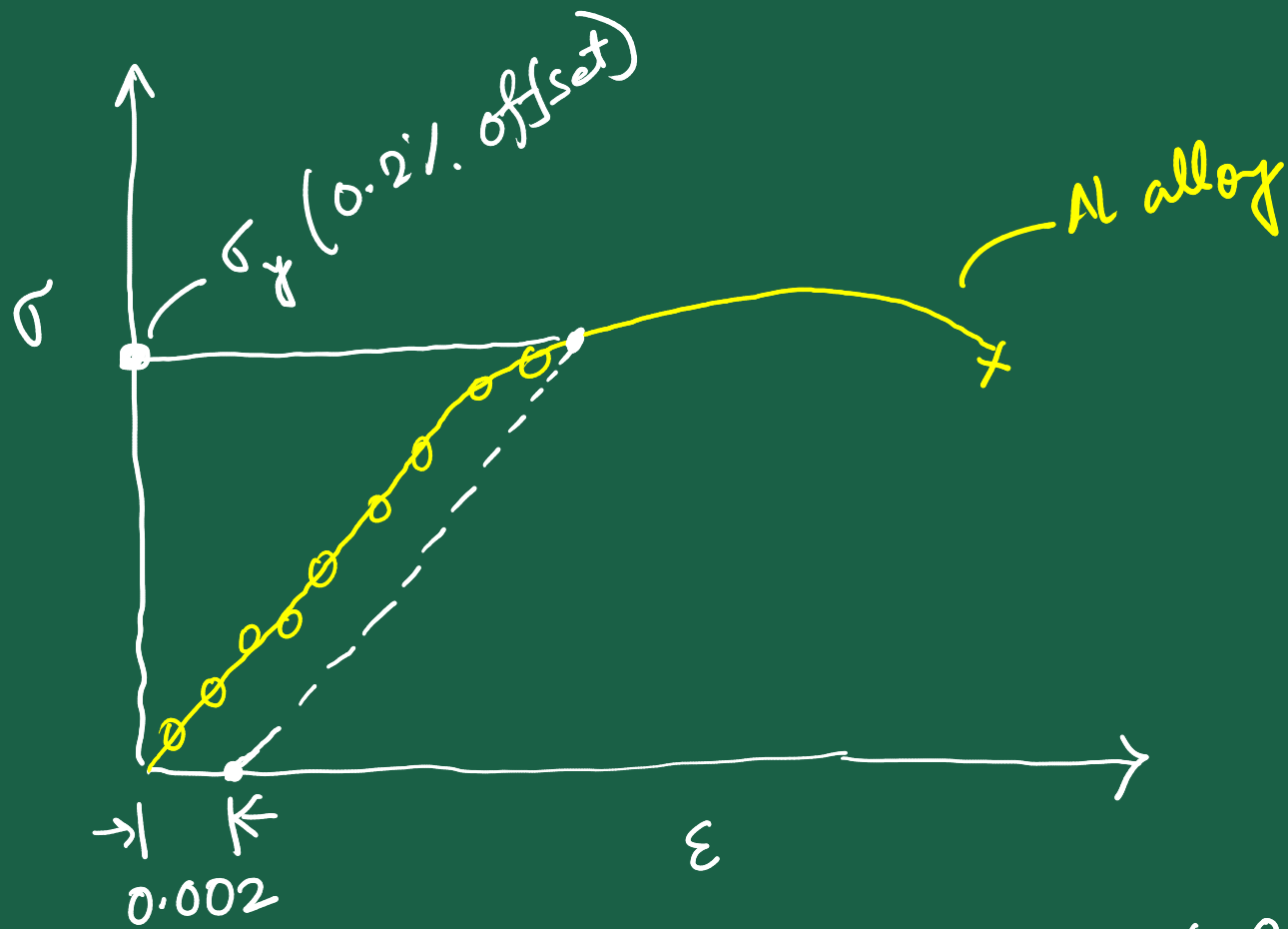
$$A < A_0 \implies \sigma_t > \sigma_e$$



$$E_{Al} < E_{St}$$

$\sim 70 \text{ GPa}$ $\sim 200 \text{ GPa}$





σ_y
 \rightarrow
 Yield

Offset : 0.2 % strain $\rightarrow \epsilon = 0.002$
 0.4 % " $\rightarrow \epsilon = 0.004$

